

$t/c = 12\%$  at  $x/c = 30\%$

- 2218 (20,e) at  $2 \cdot 10^7$
- 0015 (22,d) at  $2 \cdot 10^6$
- 0015 (20,m) at  $3 \cdot 10^6$

Figure 13. Drag coefficients (at  $C_L = 0$ ) of a family of symmetrical foil sections (with maximum thickness at 30% of the chord); (a) tested in a large-size wind-tunnel (20,a), and (b) calculated as per equation 18 for  $C_f = 0.00285 = \text{constant}$ .

0015 at transonic speeds, see (36,e) in Chapter XVII.



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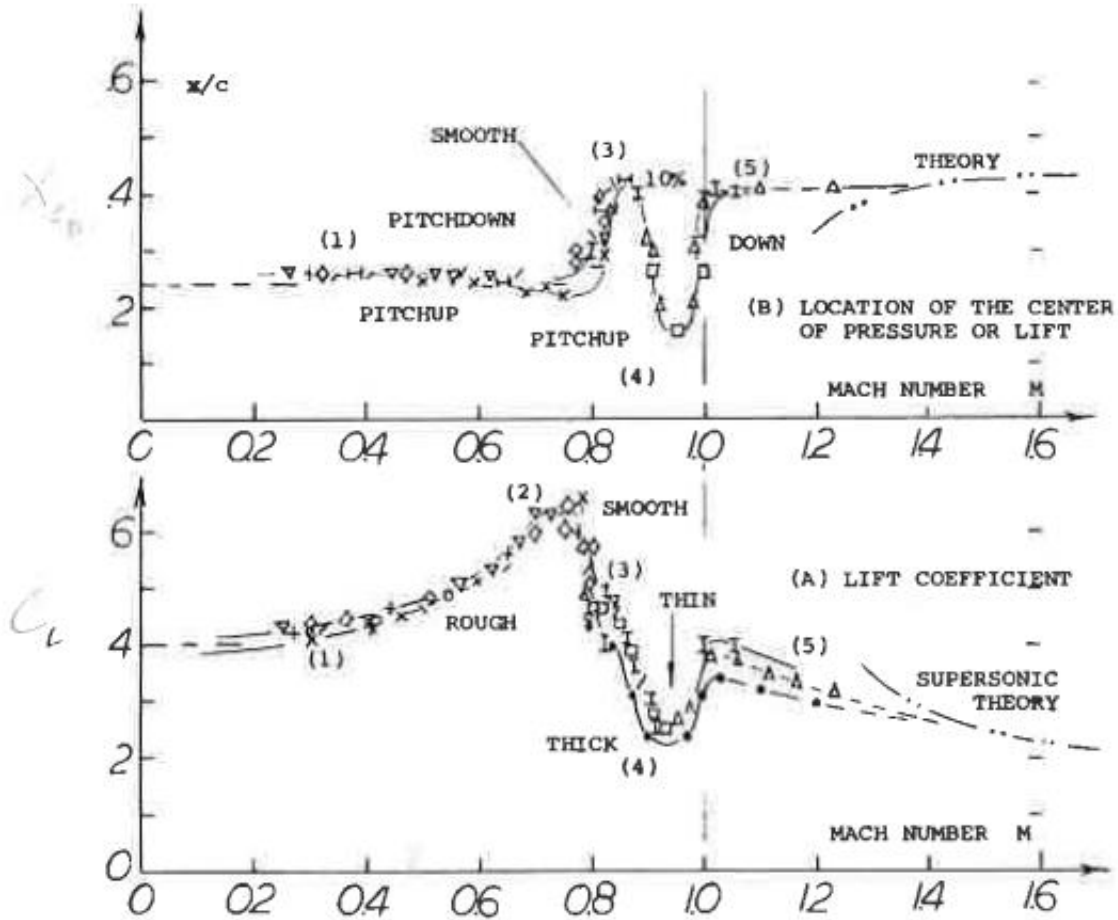


Figure 16. Lift and center of pressure of various symmetrical foil sections from subsonic through transonic M'numbers, at constant angle of attack  $\alpha = 4^\circ$ .

